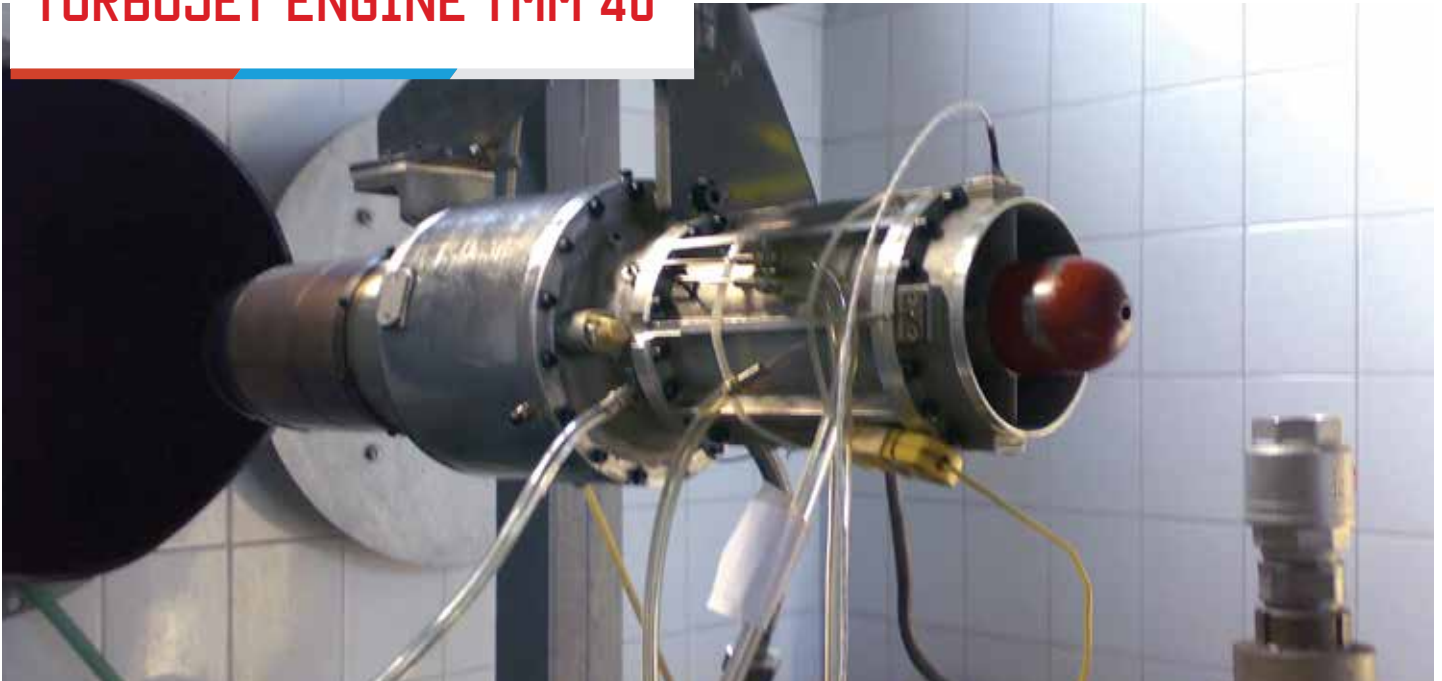
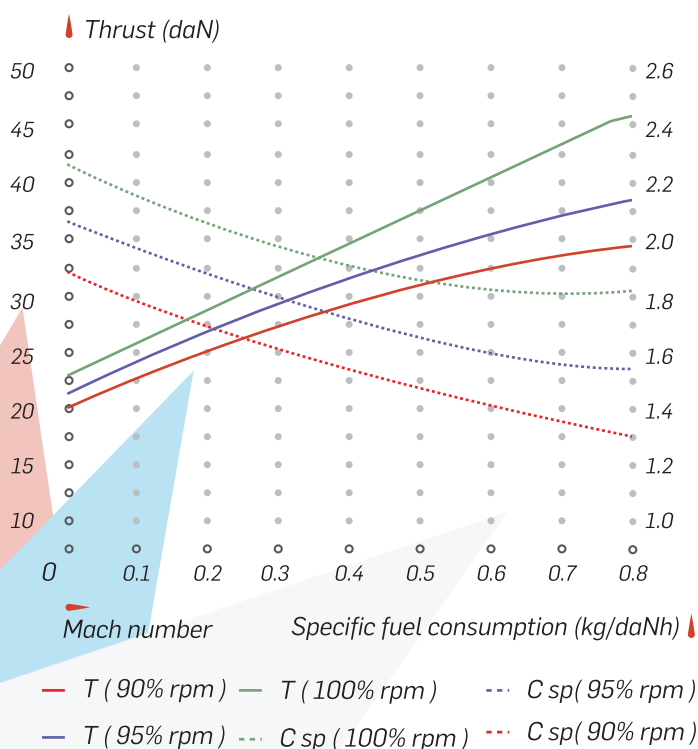


TURBOJET ENGINE TMM 40

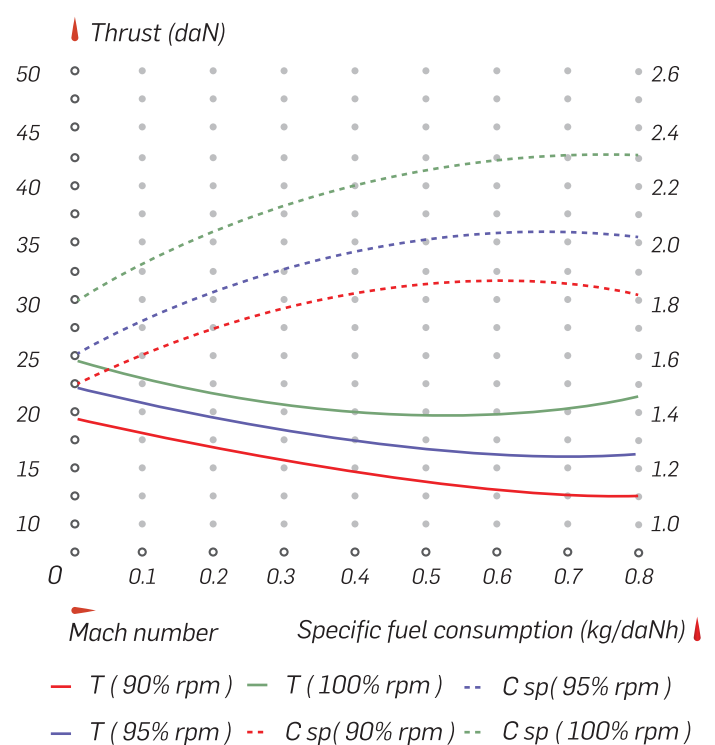


TMM40 is a small expendable turbojet engine rated for 40daN of static thrust. It has been designed to meet the low-cost engine approach, but to keep thrust-to-weight and thrust-to- cross section ratio at a competitive level. Also, the design goal was that the engine could be easily resizable to bigger engines, up to 500daN. The engine consists of three stage axial compressor, annular combustion chamber with six injectors and axial turbine. Intake and exhaust nozzle modules can be modified for certain missile designs.

Sea level static



H = 5000 m



OVERALL DIMENSIONS AND WEIGHT

Weight (dry engine)	5.6 kg
Size (mm) LxD	560x155

ISA SEA LEVEL CONDITIONS

Maximum static thrust	40 daN
Specific fuel consumption	1.42 kg/daNh
Minimum engine life	10 h

OPERATING ENVELOPE

Mach number	0/0.8
Altitude	0/5000 m
Max. rotor speed	90000 rpm

FUEL SYSTEM

Missile fuel system	
Fuel specification	MIL-T-5624

OIL SYSTEM

Air/oil mist system or self lubricated bearings with grease.	
Oil specification	MIL-L-7808

STARTING SYSTEM

Windmilling	
Pyrotechnic cartridge	

IGNITION

Pyrotechnic ignitor	
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